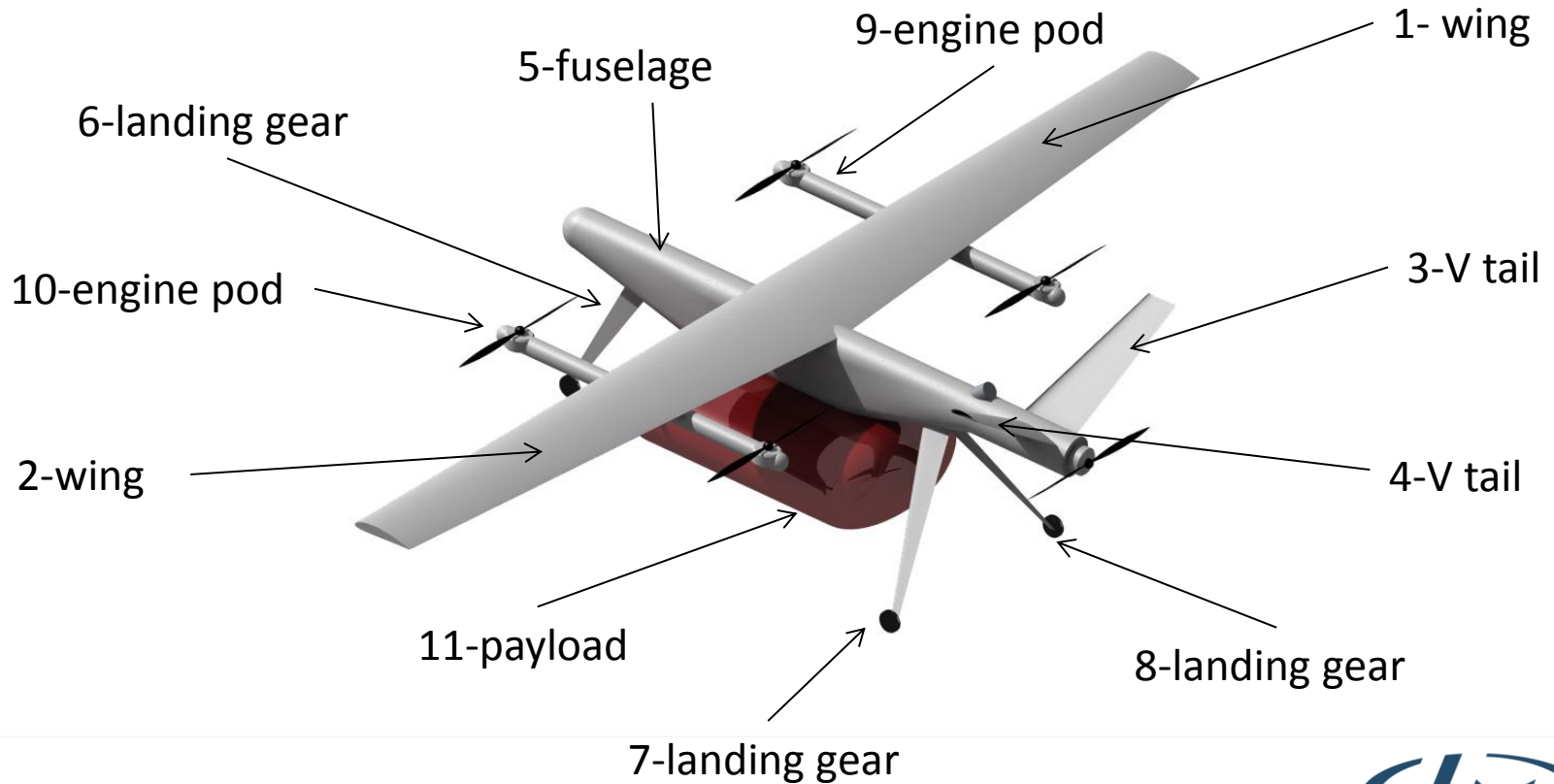


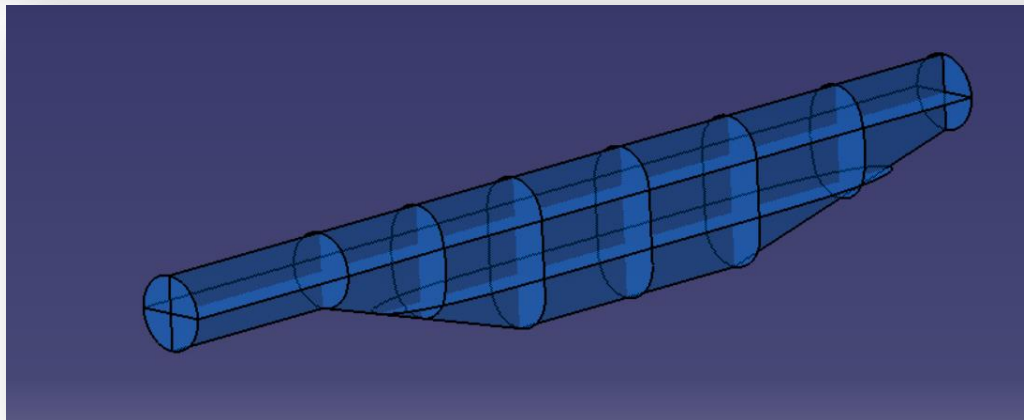
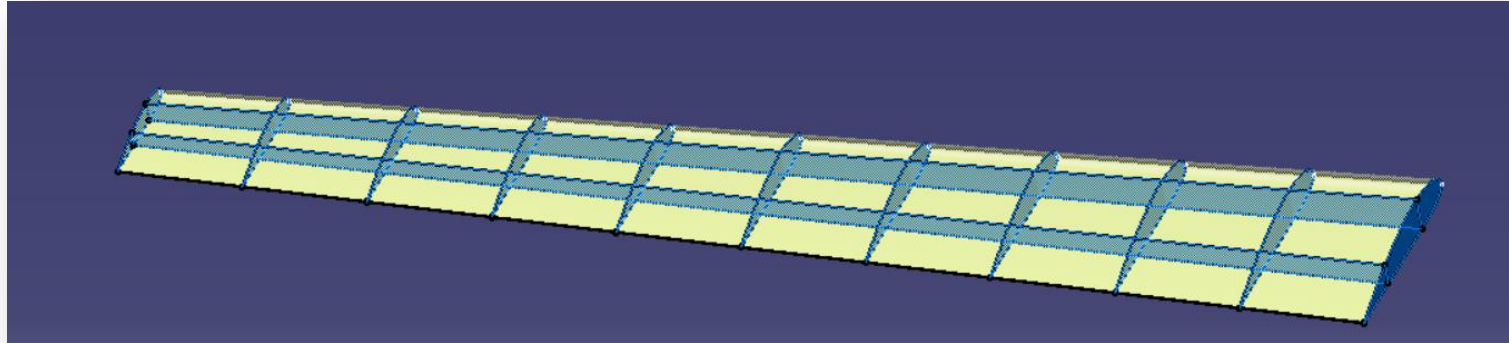
## 08.1.- STRUCTURAL LAYOUT

**Structure shall be as integrated as possible:** Carbon Fiber Wing Skins with Co-cured Spars and a similar configuration for the fuselage, integrating shear ties in the skins for frames attachment.



## 08.2.- STRUCTURAL LAYOUT – INNER STRUCTURE

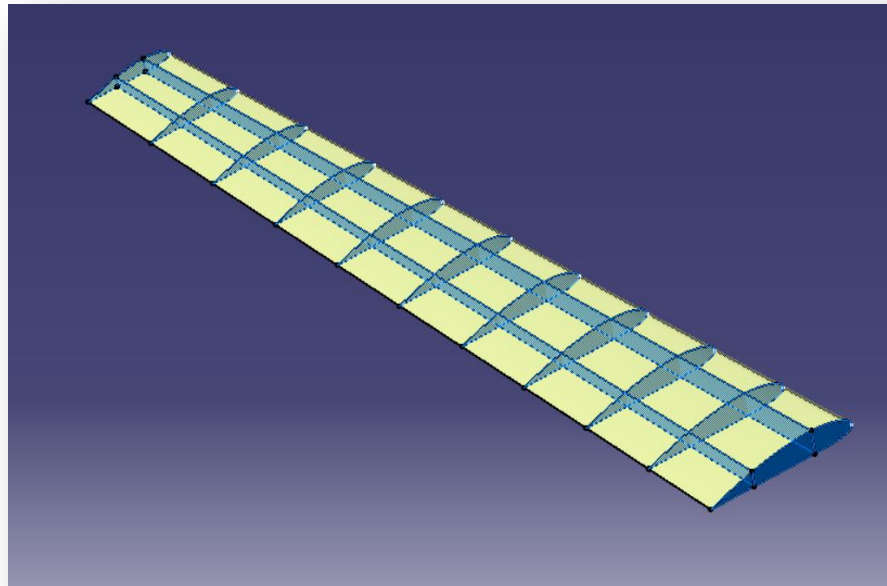
- Wings: Classical structure based on two spars and ribs
- Fuselage: Low cost carbon fiber skins, frames and stringers.



## 08.3.- STRUCTURAL LAYOUT – INNER STRUCTURE - WINGS

### WINGS

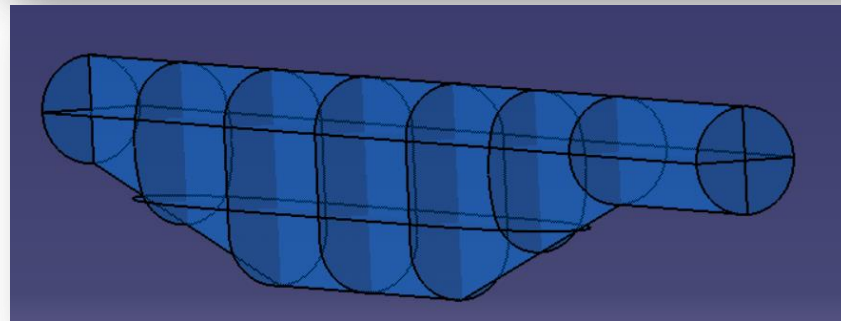
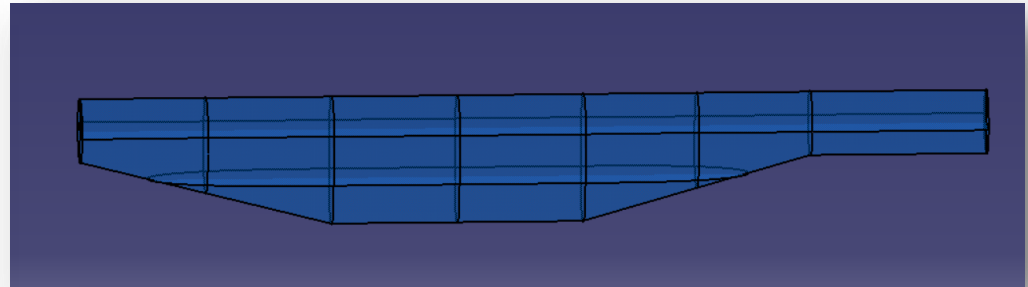
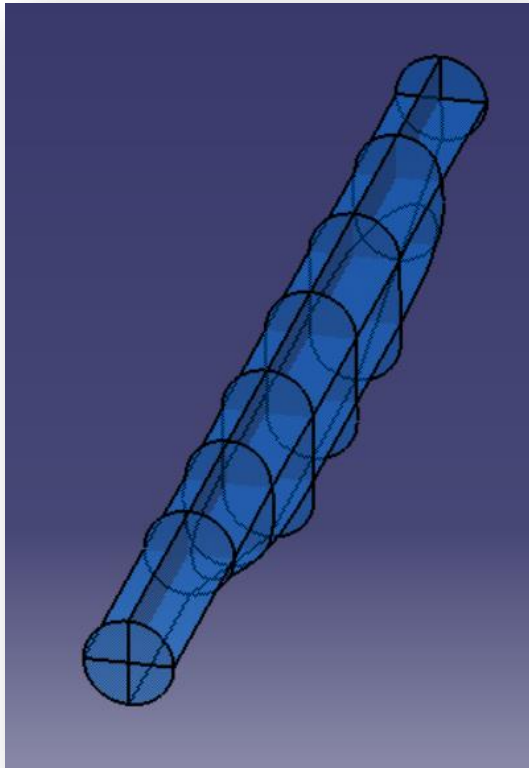
- Low cost (out of autoclave or similar) carbon fiber composite skins with integrated spars.
- Low cost carbon fiber composite intermediate ribs (riveted to skins).
- Low density aluminum (Al-Li) metallic root rib and tip rib.



## 08.4.- STRUCTURAL LAYOUT – INNER STRUCTURE - FUSELAGE

### FUSELAGE

- Low cost carbon fiber skins, frames and stringers.
- Selection of aluminum parts (to be defined during detailed Design).



## 09.- PAYLOAD DIMENSIONS

- Cargo bay/box is large enough to fit a 450x350x200mm box.

